

ABSTRACT

A gas turbine engine (10) is provided with a stage of turbine blades (22) surrounded by hollow shroud segments (28). Cooling of the shroud segments (28) is achieved by providing a flow of engine leakage air from space volume (48) centrally of the engine (10) via an annular compartment (46) wherein it is diverted away from the blade cooling system and passed via piping (44) within each guide vane (20) to shroud segments (28). The flow impingement cools the shroud segments (28) and then passes into the gas annulus via slots (60).